



BSI Standards Publication

**Space systems — Space batteries — Guidelines for
in-flight health assessment of lithium-ion batteries**

National foreword

This Published Document is the UK implementation of ISO/TR 20891:2020.

The UK participation in its preparation was entrusted to Technical Committee ACE/68, Space systems and operations.

A list of organizations represented on this committee can be obtained on request to its committee manager.

This publication does not purport to include all the necessary provisions of a contract. Users are responsible for its correct application.

© The British Standards Institution 2020
Published by BSI Standards Limited 2020

ISBN 978 0 580 92088 2

ICS 49.140

Compliance with a British Standard cannot confer immunity from legal obligations.

This Published Document was published under the authority of the Standards Policy and Strategy Committee on 31 October 2020.

Amendments/corrigenda issued since publication

Date	Text affected
------	---------------

**Space systems — Space batteries
— Guidelines for in-flight health
assessment of lithium-ion batteries**

*Systèmes spatiaux - Batteries spatiales - lignes directrices pour
l'évaluation en vol de la santé des batteries lithium-ion*





COPYRIGHT PROTECTED DOCUMENT

© ISO 2020. Published in Switzerland

All rights reserved. Unless otherwise specified, no part of this publication may be reproduced or utilized otherwise in any form or by any means, electronic or mechanical, including photocopying, or posting on the internet or an intranet, without prior written permission. Permission can be requested from either ISO at the address below or ISO's member body in the country of the requester.

ISO copyright office
Ch. de Blandonnet 8 • CP 401
CH-1214 Vernier, Geneva, Switzerland
Tel. +41 22 749 01 11
Fax +41 22 749 09 47
copyright@iso.org
www.iso.org

Contents

Page

Foreword	iv
Introduction	v
1 Scope	1
2 Normative references	1
3 Terms, definitions and abbreviated terms	1
3.1 Term and definitions.....	1
3.2 Abbreviated terms.....	1
4 Overview	2
4.1 General.....	2
4.2 Battery capacity.....	4
4.3 Battery impedance.....	4
4.3.1 General.....	4
4.3.2 Electrochemical impedance spectroscopy (EIS).....	5
4.4 Battery internal resistance.....	8
5 Specificities of spacecraft telemetry and resulting limitations	11
5.1 General.....	11
5.2 Signal digitization.....	11
5.3 Temperature.....	12
5.4 Voltage.....	12
5.5 Current.....	12
5.6 Sampling frequency.....	13
5.7 Synchronisation.....	13
5.8 On-board memory.....	13
6 Main methods for the evaluation of battery ageing parameters	14
6.1 Global method: fitting of a numerical model to in-flight data.....	14
6.1.1 General.....	14
6.1.2 Model structure.....	14
6.1.3 Data fitting.....	16
6.2 Evaluation of battery capacity.....	25
6.2.1 Direct method.....	25
6.2.2 Indirect method.....	30
6.3 Measurement of battery internal resistance.....	32
6.3.1 Direct internal resistance measurement.....	32
6.3.2 Indirect measurement of battery resistance.....	34
6.3.3 Correlation of internal resistance to capacity.....	46
6.4 Measurement of battery spectral impedance.....	47
6.4.1 General.....	47
6.4.2 Time domain identification of a dynamic model.....	48
6.4.3 Derivation of impedance from frequency domain processing of transients.....	49
6.4.4 Derivation of impedance from frequency domain processing of disturbances.....	50
7 Recommendations for easing battery in-flight health assessment	51
7.1 General recommendations.....	51
7.2 Recommendations related to battery characterization prior to flight.....	51
7.3 Recommendations related to spacecraft telemetry performance.....	52
7.4 Recommendations related to spacecraft operations.....	52
7.5 Recommendations related to data formatting.....	52
7.6 Recommendations related to data processing.....	52
Bibliography	53

Foreword

ISO (the International Organization for Standardization) is a worldwide federation of national standards bodies (ISO member bodies). The work of preparing International Standards is normally carried out through ISO technical committees. Each member body interested in a subject for which a technical committee has been established has the right to be represented on that committee. International organizations, governmental and non-governmental, in liaison with ISO, also take part in the work. ISO collaborates closely with the International Electrotechnical Commission (IEC) on all matters of electrotechnical standardization.

The procedures used to develop this document and those intended for its further maintenance are described in the ISO/IEC Directives, Part 1. In particular, the different approval criteria needed for the different types of ISO documents should be noted. This document was drafted in accordance with the editorial rules of the ISO/IEC Directives, Part 2 (see www.iso.org/directives).

Attention is drawn to the possibility that some of the elements of this document may be the subject of patent rights. ISO shall not be held responsible for identifying any or all such patent rights. Details of any patent rights identified during the development of the document will be in the Introduction and/or on the ISO list of patent declarations received (see www.iso.org/patents).

Any trade name used in this document is information given for the convenience of users and does not constitute an endorsement.

For an explanation of the voluntary nature of standards, the meaning of ISO specific terms and expressions related to conformity assessment, as well as information about ISO's adherence to the World Trade Organization (WTO) principles in the Technical Barriers to Trade (TBT), see www.iso.org/iso/foreword.html.

This document was prepared by Technical Committee ISO/TC 20, *Aircraft and space vehicles*, Subcommittee SC 14, *Space systems and operations*.

Any feedback or questions on this document should be directed to the user's national standards body. A complete listing of these bodies can be found at www.iso.org/members.html.

Introduction

The charge and discharge cycle of a battery is not 100 % efficient, with each cycle side reactions can occur that eventually accumulate and cause degradation of the battery's performance. Understanding how the battery's performance changes throughout the mission is a subject of importance; and accurate determination of the battery's current SoH is essential in a large number of situations, for example:

- the routine assessment of battery performance to allow early detection of anomalies (by comparing its actual versus predicted performance);
- the setting of alarm thresholds to ensure adequate energy;
- detection of battery anomalies that can put at risk the spacecraft passivation and/or de-orbiting strategy;
- decisions regarding mission extension beyond initial target life;
- evaluating the remaining capability of a spacecraft upon occurrence of an anomaly;
- feedback to the battery manufacturer to improve the performance prediction.

However, it is often difficult to properly assess the in-flight status, due to various factors:

- Flight electrical load profiles differ significantly to load profile used to characterize battery performance models and the battery's SoH; for example, the total available battery capacity, which is the most important parameter, is not directly accessible during flight since its simple measurement by full discharge of the battery goes against the spacecraft operational safety.
- The quality of the accessible data from telemetry is sometimes poor: insufficient telemetry resolution and/or accuracy, lack of synchronization between related parameters like current and voltage, possibly large load consumption fluctuations introducing a high level of noise, delivery of data under a form not easy to process, etc.
- The battery is operating in flight in a way that is generally very different from the test conditions at qualification or acceptance. As a consequence, if no in-flight assessment has been made at the beginning of life, the direct comparison between current in-flight status and available ground testing data can be difficult and in any case more difficult than a comparison with the initial in-flight behaviour.
- The battery is operated under time variant conditions in a large bandwidth of different time scales, e.g. switching heater circuits vs. variations of the charge profile and eclipse length for a LEO satellite with drifting orbit. Low frequency variations introduced by drifting orbits or seasons are considered for the computation of trends and averaging over several orbits.
- The processing of data to derive the health status is not straightforward and is usually performed by identifying the ageing parameters of a model. Therefore, the representativeness of this model is a key issue. In addition, even with a good model, the results are not always satisfactory.

Therefore, it has been found of interest to provide detailed information about the means currently used or envisioned to perform in-flight battery health assessment and to make recommendations to spacecraft builders, operation managers and batteries manufacturers that would make it easier. This is the subject of this document.

It is important to highlight that, according to the definition given in [3.1.1](#), assessing the health status allows to verify that the battery behaves as well as or possibly better than anticipated. It is not aimed at providing an evaluation of any sort of "absolute ageing" nor to predict further evolution, even if this can be the case with some methods and their on-board implementation.

Space systems — Space batteries — Guidelines for in-flight health assessment of lithium-ion batteries

IMPORTANT — The electronic file of this document contains colours which are considered to be useful for the correct understanding of the document. Users should therefore consider printing this document using a colour printer.

1 Scope

This document provides detailed information on the various methods of assessing the health status of lithium-ion space batteries in flight and makes recommendations to battery suppliers, spacecraft manufacturers and operators to ease this assessment.

2 Normative references

The following documents are referred to in the text in such a way that some or all of their content constitutes requirements of this document. For dated references, only the edition cited applies. For undated references, the latest edition of the referenced document (including any amendments) applies.

ISO 17546, *Space systems — Lithium ion battery for space vehicles — Design and verification requirements*

3 Terms, definitions and abbreviated terms

3.1 Term and definitions

For the purposes of this document, the terms and definitions given in ISO 17546 and the following apply.

ISO and IEC maintain terminological databases for use in standardization at the following addresses:

- ISO Online browsing platform: available at <https://www.iso.org/obp>
- IEC Electropedia: available at <http://www.electropedia.org/>

3.1.1

battery health

state of the battery, which is healthy if both the amount and the rate of degradation of its performance are lower than or equal to the predicted ones at the same time into the mission

3.2 Abbreviated terms

ADC	analogue to digital converter
BOL	beginning of life
CC	constant current
CV	constant voltage
DoD	depth of discharge
EIS	electrochemical impedance spectroscopy
EMF	electro-motive force (a.k.a. open circuit voltage)